Airplane Weight and Balance

Builder: _____ Date: _____
Model: _____ Registration: _____
Gross weight: _____ pounds (lbs.) Serial #: _____
CG Range: _____ inches (in.)
Datum is: _____

Empty Weight & CG	Wt. (lb.)	Arm (in.)	Moment (in.lb.)
Left wheel =			0
Right wheel =			0
Nose/Tail =			0
Aircraft empty weight =	**0**		0
Empty CG =		**#DIV/0!**	

Most Aft Weight & CG	Wt. (lb.)	Arm (in.)	Moment (in.lb.)
Aircraft Empty =	0		0
Pilot Seat =			0
passenger Seat =			0
Baggage =			0
Fuel =			0
Weight =	**0**		0
CG =		**#DIV/0!**	

Most Forward Weight & CG	Wt. (lb.)	Arm (in.)	Moment (in.lb.)
Aircraft Empty =	0		0
Pilot Seat =			0
Passenger Seat =			0
Baggage =			0
Fuel =			0
Weight =	**0**		0
CG =		**#DIV/0!**	

Flight Test Weight & CG	Wt. (lb.)	Arm (in.)	Moment (in.lb.)
Aircraft Empty =	0		0
Pilot Seat =			0
Passenger Seat =			0
Fuel =			0
Weight =	**0**		0
CG =		**#DIV/0!**	

Notes:

1. The most forward and most aft CG limits are calculated using the FAA standard pilot and passenger weight of 170 pounds.
2. Minimum fuel weight is calculated using the formula: ((Max. continuous hp) / 12) X 6
3. For most aft CG calculation maximize all weight aft of the aft CG limit and minimize all weights forward of the aft CG limit.
4. For most forward CG calculations maximize all weight forward of the forward CG limit and minimize all weights aft of the forward CG limit.
5. Pilot of 170 pounds must be in both Most Aft and Most Forward CG calculations.
6. Weigh aircraft in a level flight attitude.
7. Weight X Arm = Moment
8. Moment / Weight = CG